

Amendments to the Claims:

This listing of claims will replace all prior versions, and listings, of claims in the application:

Listing of Claims:

1-14 (cancelled)

15. (currently amended) A gas turbine engine fuel injection and combustor system, comprising:

an outer casing extending from an upstream end to a downstream end, an internal space of the downstream end defining a combustion chamber;

5 an annular dome connected to an internal, upstream end of the outer casing;

a radial flow air swirler mounted to an internal surface of the annular dome and to an external surface of a fuel injector body, said swirler providing swirled air to the combustion chamber; and

10 a fuel injector body mounted to an internal surface of the radial flow air swirler, said fuel injector body comprising a plurality of axially oriented air swirlers, an axially located pilot circuit fuel nozzle, and a plurality of radially oriented main circuit fuel nozzles; and

wherein the annular dome includes a plurality of circumferentially disposed dome cooling nozzles; and

wherein said fuel injector body incorporates both the pilot circuit and the main circuit fuel nozzles.

16. (cancelled)

17. (currently amended) The gas turbine engine fuel injection and combustor system of claim 15, wherein said fuel injector body for the pilot and main circuit fuel ~~injection systems nozzles~~ is cylindrical.

18. (cancelled)

19. (currently amended) The gas turbine engine fuel injection and combustor system of claim 15, wherein said pilot circuit fuel nozzle is encircled by ~~a~~ the plurality of axial axially oriented air swirlers.

20. (currently amended) The gas turbine engine fuel injection and combustor system of claim 15, wherein said main circuit fuel nozzles each discharge fuel ~~positioned at a compound an angle with respect to a radially extending axial axis of each nozzle~~ into a radial swirler passage.

21. (cancelled)

22. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, further comprising a can type combustor.

23. (previously presented) The gas turbine engine fuel injection and combustor system of claim 15, further comprising an annular type combustor.

24. (currently amended) The gas turbine engine fuel injection and combustor system of claim 15, ~~comprising~~ wherein the system is capable of utilizing aviation fuel.

25. (currently amended) A gas turbine engine fuel injection and combustor system, comprising:

an outer casing extending from an upstream end to a downstream end, an internal space of the downstream end defining a combustion chamber;

5 an annular dome connected to an internal, upstream end of the outer casing;

 a radial flow air swirler mounted to an internal surface of the annular dome and to an external surface of a fuel injector body, said swirler providing swirled air to the combustion chamber; and

10 a fuel injector body mounted to an internal surface of the radial flow air swirler, said fuel injector body comprising a plurality of axially oriented air swirlers, an axially located pilot circuit fuel nozzle, and a plurality of radially oriented main circuit fuel nozzles;

15 wherein the combined geometry of the pilot circuit and main circuit fuel injection systems provides a combined volume that is less than the volume of the combustion chamber

wherein said fuel injector body incorporates both the pilot circuit and the main circuit fuel nozzles.

26. (cancelled)

27. (currently amended) The gas turbine engine fuel injection and combustor system of claim 25, wherein said fuel injector body for the pilot and main circuit fuel ~~injection systems~~ nozzles is cylindrical.

28. (cancelled)

29. (currently amended) The gas turbine engine fuel injection and combustor system of claim 25, wherein said pilot circuit fuel nozzle is encircled by a the plurality of axial axially oriented air swirlers.

30. (currently amended) The gas turbine engine fuel injection and combustor system of claim 25, wherein said main circuit fuel nozzles each discharge fuel positioned at a compound angle with respect to a radially extending axial axis of each nozzle into a radial swirler passage.

31. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, further comprising a can type combustor.

32. (previously presented) The gas turbine engine fuel injection and combustor system of claim 25, further comprising an annular type combustor.

33. (currently amended) The gas turbine engine fuel injection and combustor system of claim 25, comprising wherein the system is capable of utilizing aviation fuel.

34-42. (cancelled)